

1181. A research on the dynamic characteristics of axially moving Timoshenko beam with compressive load

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Abstract. The free vibration characteristics of the axially moving Timoshenko beam under compressive load are investigated. The differential governing equation of transverse vibration of the axially moving beam under axial load is established based on the Timoshenko beam theory and Hamilton's principle. The beam system is sorted by the positive definiteness of boundary conditions and the characteristic equations are deduced. The dimensionless complex frequencies of the axially moving Timoshenko beam with different boundary conditions under compressive load are calculated by the proposed modified Galerkin method and the differential quadrature method. The characteristics of the lower order real parts and imaginary parts of the complex frequencies are observed. The effect of compressive load factor, the moving speed and acceleration on the stability of system are discussed.

Keywords: axially moving Timoshenko beam, compressive load, positive definiteness, transverse vibration.

1. Introduction

Axially moving continuum is involved in industry of engineering and aerospace, such as bandsaws in engineering devices, magnetic bands and moving rockets with high speed, missiles and rigid pipeline in aircrafts and some other components, etc. Most of those devices can be simplified as axially moving beams. Dues to effects of axially moving and excitations of axial load, longitudinal and torsional vibrations can be ignored in contrast with transverse vibration of those systems. Therefore, Study of transverse vibration and dynamic behavior of axially moving beam is of great significance for performance improving and optimization design for these systems.

Many researches have been carried out on axially moving beam. Swope and Ames [1] established a mathematical model for the oscillation of a string as it is traversed and wound on a bobbin, and noted the importance of the ratio of the winding speed to wave velocity. Mote [2] found that the flexural natural frequencies of bandsaw decrease continuously with increasing band velocity and large amplitude equilibrium may be induced by small periodic band tension variations as in the case of the classical string. Lengoc [3] studied the free vibrations of axially moving bandsaws by eigenvalue method and its stability under parametric excitations. Wickert [4] analyzed the free non-linear vibration of axially moving tensioned beam in sub and supercritical transport speed ranges for the pinned-pinned boundary condition. Öz [5, 6] calculated the natural frequencies for different flexural stiffness with time varying velocity using multi-scale approach. Kong and Parker [7] used perturbation method to obtain algebraic equations of natural frequencies and studied the free vibration of the axially moving beam with small flexural stiffness. Yang and Chen [8] also used multi-scale approach to determin the natural frequencies of axially moving beam. Lee and Jang [9] studied the dynamic characteristics and stability of an axially moving beam with infinite parts under different boundary conditions by using the spectral element method.

Most of these beam models are based on the Euler-Bernoulli beam theory, and it appears not so accurate for beams with low length-thickness ratio as a result of without considering shear deformation and rotary inertia of cross section. Simpson [10] firstly investigated the shear effect and analyzed the natural frequencies of the axially moving Timoshenko beam with clamped-clamped boundary condition, in which the eigenvalue method was used and the

divergence of frequency was shown. Chonan [11] used Laplace transform method to analyze the steady-state response of axially moving Timoshenko beam.

However, no axial load was considered in above two beam models. Lee et al. [12] formulated the spectral element model for the axially moving Timoshenko beam under a uniform axial tension and used exact dynamic stiffness matrix to provide high accuracy of solutions. By applying complex mode approach, Tang et al. [13] analyzed natural frequencies, modes and critical speeds of axially moving Timoshenko beams on different supports under uniform tension. On the other hand, there are some works for static beam with compressive axial load. Bokaina [14, 15] had investigated axial tension and compressive load effect for Euler-Bernoulli beam; Zhang et al. [16] studied the effect of compressive axial load on transverse vibrations of a double-beam system and found that the properties of the forced transverse vibrations of the system are significantly dependent on the compressive axial load. Christian and Shu [17] analyzed the influence of axial compressive load on the vibration characteristics of a multiple-delaminated beam. These are all for static beam without considering axially moving effect, and the boundary conditions of support are investigated for clamped and pinned situations, which is belong to the positive definite system (PDS), but for the positive semi-definite system (PSDS), such as free-free boundary conditions, there were few researches. Pourtakdoust and Assadian [18] investigated the thrust effect on the vibrational characteristics of flexible guided missiles, but axially motion was not involved. Therefore, to date no publication can be found for the study of moving beam with compressive load, which is still of key engineering significance.

In this paper, the governing equation of transverse vibration of the axially moving beam under compressive axial load is established based on the Timoshenko beam theory and Hamilton's principle at first, then the dimensionless complex frequencies of the axial moving Timoshenko beam with different boundary conditions of PDS and PSDS are calculated numerically by the proposed modified Galerkin method and the differential quadrature method (DQM). The effect of compressive load factor, the moving speed and acceleration on the stability of system are analyzed, and the critical speeds and loads on different boundary conditions are predicted.

2. Governing equations

For an axially moving Timoshenko beam travelling with speed $V(T)$ under axial compressive constant load N , the stiffness of its cross section is set as $EI(X)$ and the shear stiffness is κGA . $I(X)$ is the moment of inertia of cross section along length and E is the modulus of elasticity. κ , G , A , L and ρ represent the shape factor, shearing modulus, area of cross section, length of beam and density of beam material, respectively. The transverse vibration can be described by the axial coordinate X and time T , and the displacements in the axial and transverse coordinates of axial and transverse are $U(X, T)$ and $W(X, T)$, respectively. For uniform beams, rotatory inertia of cross section can be represented by ρI . The model of the current moving Timoshenko beam under compressive load is shown in Fig. 1.

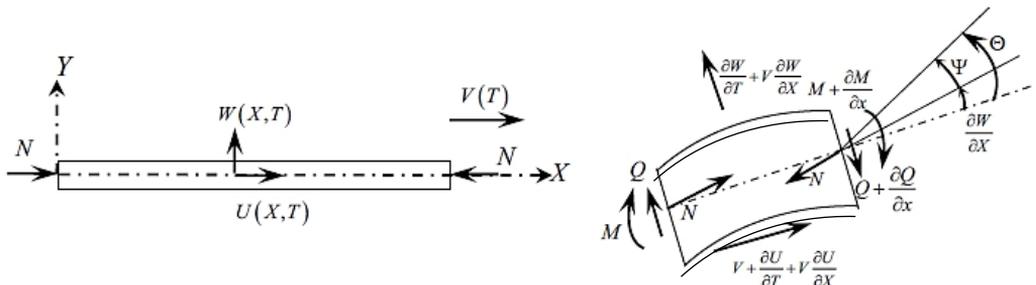


Fig. 1. Model of axially moving Timoshenko beam under compressive load

Where Θ is the angle of rotation of cross section and can be expressed as $\Theta = \partial W / \partial X + \Psi$ for Timoshenko beam, Ψ is the shear angle. For Euler-Bernoulli beam, $\Theta = \partial W / \partial X$.

Without considering the coupling of longitudinal and transverse vibrations, the kinetic energy of the beam involves transverse vibration energy, rotational kinetic energy of cross section and the kinetic energy of rigid motion:

$$T = \frac{1}{2} \int_0^L \rho A \left(\frac{\partial W}{\partial T} + V \frac{\partial W}{\partial X} \right)^2 dX + \frac{1}{2} \int_0^L \rho I \left(\frac{\partial \Theta}{\partial T} \right)^2 dX + \frac{1}{2} \int_0^L \rho A V^2 dX. \quad (1)$$

The components of potential energy involves potential energy due to compressive load, and strain energy of bending and shear:

$$U = \frac{1}{2} \int_0^L -N \left(\frac{\partial W}{\partial X} \right)^2 dX + \frac{1}{2} \int_0^L EI \left(\frac{\partial \Theta}{\partial X} \right)^2 dX + \frac{1}{2} \int_0^L \kappa GA \left(\Theta - \frac{\partial W}{\partial X} \right)^2 dX. \quad (2)$$

According to Hamilton's principle, the difference between the kinetic and potential energy is zero:

$$\delta \int_0^{T_1} (T - U) dT = 0. \quad (3)$$

Substituting Eqs. (1) and (2) into Eq. (3), the expression of governing equations for axially moving Timoshenko beam under compressive load can be established based on the variational analysis in terms of axially moving speed and acceleration, but without derivative to load with respect to the axial coordinate and time:

$$\rho A \left[\frac{\partial^2 W}{\partial T^2} + 2V \frac{\partial^2 W}{\partial X \partial T} + V^2 \frac{\partial^2 W}{\partial X^2} + \frac{\partial V}{\partial T} \frac{\partial W}{\partial X} \right] + N \frac{\partial^2 W}{\partial X^2} + \frac{\partial}{\partial X} \left[\kappa GA \left(\Theta - \frac{\partial W}{\partial X} \right) \right] = 0, \quad (4)$$

$$\kappa GA \left(\Theta - \frac{\partial W}{\partial X} \right) - \frac{\partial}{\partial X} \left(EI \frac{\partial \Theta}{\partial X} \right) + \frac{\partial}{\partial T} \left(\rho I \frac{\partial \Theta}{\partial T} \right) = 0, \quad (5)$$

where:

$$\frac{\partial \Theta}{\partial X} = - \frac{\rho A}{\kappa GA} \left[\frac{\partial^2 W}{\partial T^2} + 2V \frac{\partial^2 W}{\partial X \partial T} + V^2 \frac{\partial^2 W}{\partial X^2} + \frac{\partial V}{\partial T} \frac{\partial W}{\partial X} \right] - \frac{1}{\kappa GA} N \frac{\partial^2 W}{\partial X^2} + \frac{\partial^2 W}{\partial X^2}. \quad (6)$$

Substituting Eq. (4) into Eq. (5) yields the partial differential equations which contains transverse vibration only:

$$\begin{aligned} W_{TT} + 2VW_{XT} + V^2W_{XX} + \frac{\rho I}{\kappa GA} W_{TTTT} + V_T W_X + \frac{N}{\rho A} W_{XX} + \frac{EI}{\kappa GA} V_T W_{XX} + \frac{2\rho IV}{\kappa GA} W_{XTT} \\ + \frac{\rho I}{\kappa GA} (2V_T^2 W_{XX} + 4VV_T W_{XXT} + 5V_T W_{XTT}) - \left(\frac{EI}{\kappa GA} + \frac{I}{A} - \frac{NI}{\kappa GA^2} - \frac{\rho I}{\kappa GA} \right) W_{XXTT} \\ - \frac{2EIV}{\kappa GA} W_{XXXT} + \left(\frac{EI}{\rho A} - \frac{NEI}{\kappa G \rho A^2} - \frac{EI}{\kappa GA} V^2 \right) W_{XXXX} = 0. \end{aligned} \quad (7)$$

By introducing the dimensionless parameters:

$$x = \frac{X}{L}, \quad w = \frac{W}{L}, \quad t = T \sqrt{\frac{N}{\rho AL^2}}, \quad v_t = V_T \frac{\rho AL}{N},$$

$$k_1 = \frac{EI}{\kappa GAL^2}, \quad k_2 = \frac{NI}{\kappa GA^2 L^2}, \quad k_3 = \frac{I}{AL^2}, \quad k_4 = \frac{EI}{NL^2}, \quad k_5 = \frac{N}{\kappa GA}.$$

Equation (7) can be expressed as dimensionless form:

$$\begin{aligned} v_t w_x + w_{tt} + 2v w_{xt} + (2k_2 v_t^2 + v^2 + 1)w_{xx} + 4k_2 v v_t w_{xxt} + 5k_2 v_t w_{xtt} + k_1 v_t w_{xxx} \\ + (k_4 - k_1 - k_1 v^2)w_{xxxx} - 2k_1 v w_{xxx} - (k_1 + k_3 - k_2 - k_2 v^2)w_{xxtt} \\ + 2k_2 v w_{xtt} + k_2 w_{ttt} = 0, \end{aligned} \quad (8)$$

where the dimensionless parameters k_1 , k_2 and k_5 associated with G account for the effect of shear distortion, k_3 and k_4 represent the effect of rotary inertia and stiffness of the beam, respectively. Bending moment $M(X)$ and shear force $Q(X)$ can be defined as $M(X) = EI \partial \theta / \partial X$, $Q(X) = M_x = \kappa GA(\theta - \partial W / \partial X)$.

Equation (8) contains 1st to 4th order partial differential components of dimensionless axial coordinate and time, and the general solution in the form of separation of variables can be expressed as:

$$w(x, t) = \phi(x) e^{i\lambda t}, \quad (9)$$

where $\phi = w(x)$ is the mode shape function, $i^2 = -1$ and λ denotes the natural frequency of beam.

The dimensionless form of different boundary conditions are shown in Table A-1 in Appendix, and the static critical loads of each kind of support under compressive load are also given. It should be noted that the loads are true value for positive definite system (pinned-pinned, clamped-clamped, clamped-free) and for positive semi-definite system (free-free and pinned-free), critical load is assumed to be equal to that of pinned-pinned boundary condition for reference. Moreover, the positive semi-definite systems are assumed to be kept at steady state like flight vehicles, rigid motion of instability is restricted.

Boundary conditions used in reference [13] is a simplified form without taking into account of the impact of time-varying characteristics, which means the coefficient of λ is zero. Though it makes the equations easy to solve, the result is not so accurate. In the present work, the items of time-varying are included in the boundary conditions as shown in Table A-1.

3. Solving methods

3.1. Analytical solution of axial compressive load effect

The governing equation of transverse vibration of Euler-Bernoulli beam under axial compressive load can be translated to a dimensionless form for static beam as [14]:

$$w_{xxxx} + w_{xx} / k_4 - \lambda^2 w = 0, \quad (10)$$

where λ is the eigenvalue which has a series of solution with the form $\lambda_i^2 = \bar{\omega}_i L^2 \sqrt{\rho A / EI}$, $i = 1, 2, \dots$, and $\bar{\omega}_i$ is the i th natural frequency of beam. w_{xx} / k_4 represents the effect of compressive load and the general solution of Eq. (10) is:

$$w(x) = C_1 \sin \alpha_1 x + C_2 \cos \alpha_1 x + C_3 \sinh \alpha_2 x + C_4 \cosh \alpha_2 x, \quad (11)$$

where C_1 , C_2 , C_3 and C_4 are constant coefficients. The characteristic equations of each boundary conditions are obtained by substituting equations in Table A-1 into Eq. (11), as shown in Table A-2 in Appendix.

Defining the variable of load $k_s = k_4^{-0.5} = \sqrt{NL^2/EI}$, we have $\alpha_1 = (0.5k_s^2 + \sqrt{0.5k_s^4 + \lambda^4})^{1/2}$ and $\alpha_2 = (-0.5k_s^2 + \sqrt{0.5k_s^4 + \lambda^4})^{1/2}$. Moreover, the dimensionless load factor is defined as the ratio of true load to critical load:

$$k_n = N/N_{cr} = (k_s^2 EI/L^2)/(\mu EI/L^2) = k_s^2/\mu, \tag{12}$$

where μ is a coefficient depending on the boundary condition, it takes π^2 for pinned-pinned, pinned-free and free-free, $\pi^2/4$ for clamped-free and $4\pi^2$ for clamped-clamped, as shown in Table A-1. There are no analytical solutions for equations in Table A-2 except pinned-pinned and pinned-free boundary conditions, for those cases the first order eigenvalue λ_1 can be obtained with the variation of axial compressive load by numerical solution, as shown in Fig. 2.

It is shown in Fig. 2 that λ_1 for clamped-clamped and free-free beams are the same. For all boundary conditions λ_1 decreases slowly with the increasing axial load in the range of $k_n = 0.01-0.8$; when the load is near to critical load as $k_n = 0.8-1$, λ_1 decreases rapidly for all boundary conditions, when the load reaches to the critical load, λ_1 decreases to 0. Thus, the axial load causes the beam stiffness decreasing and the first order frequency of elastic mode reduces to 0 when it reach to the critical load while the beam undergoes instability state. Translating the natural frequencies to dimensionless forms and the relationship between dimensionless frequencies and axial compressive load can be obtained:

$$\omega_i = \bar{\omega}_i \sqrt{\rho AL^2/N} = \lambda_i^2 / \sqrt{\mu k_n}. \tag{13}$$

It is shown in Eq. (13) that ω_i is proportional to λ_i^2 , but is inversely proportional to load $\sqrt{k_n}$, and is also dependent on factor μ of boundary condition. Fig. 3 presents the relationship between the first order dimensionless frequencies ω_1 and the load factor k_n . Since $k_n \rightarrow 0$, $\omega_i \rightarrow +\infty$, load factor is defined in the range of $k_n = 0.01-1$ in this paper. ω_1 is different from λ_1 and it decreases rapidly when $k_n = 0.01-0.1$ and more slowly when $k_n = 0.1-1$. ω_1 decreases to 0 when $k_n = 1$. The results of analytical solution can be used to verify with the numerical methods.

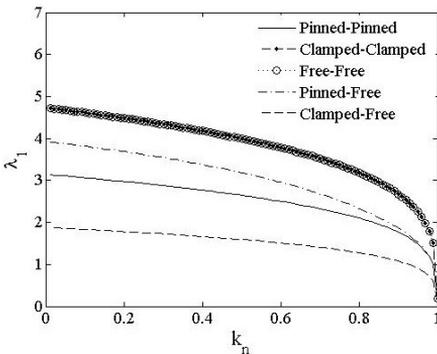


Fig. 2. λ_1 vs. k_n of different boundary conditions

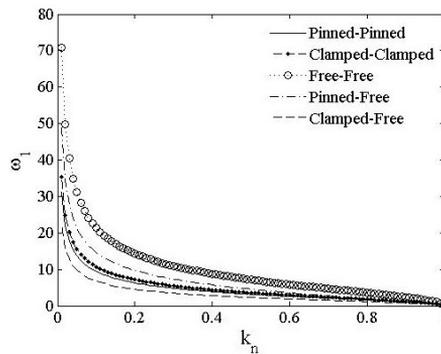


Fig. 3. ω_1 vs. k_n of different boundary conditions

3.2. Modified Galerkin method (MGM)

The Galerkin method can be used to solve Eq. (8), when its trial functions satisfy both force and displacement boundary conditions. The traditional Galerkin method takes mode functions of a static beam without axial load as its trial functions and thus leads to some inaccuracy for solving such problems. In this work, the Galerkin method is modified by taking the mode functions of beam with load as its trial functions, and thus the present algorithm is called the Modified Galerkin method (MGM). Variation of the first four order roots with load factor k_n can be obtained by

solving equations in Table A-2 numerically, as shown in Fig. 4.

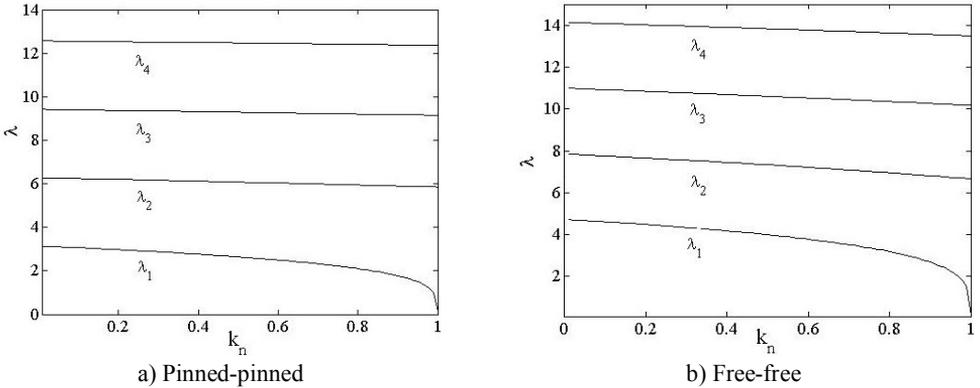


Fig. 4. First four order eigenvalues vs. load of different boundary conditions

The 2nd, 3rd and 4th order eigenvalues decrease slowly with increasing compressive load and without significant drop like the 1st order eigenvalue when the load is close to the critical load, which indicates that the axial load has little effect on the higher-order eigenvalues other than the first order. Stiffness coefficient k_4 reaches its minimum value when the axial load closes to the critical one, and the contribution of the 1st order eigenvalue caused by compressive load reaches to a maximum value. Obviously, eigenvalues of non-axial load beam are just a special state of pre-loaded beam when $k_n = 0$.

Substituting equations of boundary condition in Table A-1 into Eq. (11) yields the mode functions of beam under compressive load, as shown in Table A-3 in Appendix. Equation (9) is rewritten as $w(x, t) = \Phi^T \mathbf{q}$, $\Phi = [\phi_1(x) \ \phi_2(x) \ \dots \ \phi_n(x)]$, $\mathbf{q} = [q_1(t) \ q_2(t) \ \dots \ q_n(t)]$, $q_j(t) = e^{i\lambda_j t}$. $\phi_i(x)$ is the Mode shape functions $\phi(x)$ in Table A-3 which is used as the trial functions. From Eq. (8) we have:

$$\begin{aligned}
 k_2 \mathbf{q}_{tttt} + 2 k_2 v \mathbf{B}_1 \mathbf{q}_{ttt} + [\mathbf{I} + 5 k_2 v_t \mathbf{B}_1 - (k_1 + k_3 - k_2 - k_2 v^2) \mathbf{B}_2] \mathbf{q}_{tt} \\
 + [2v \mathbf{B}_1 + 4 k_2 v v_t \mathbf{B}_2 - 2 k_1 \mathbf{B}_3] \mathbf{q}_t \\
 + [v_t \mathbf{B}_1 + (2k_2 v_t^2 + v^2 + 1) \mathbf{B}_2 + k_1 v_t \mathbf{B}_3 + (k_4 - k_1 - k_1 v^2) \mathbf{B}_4] \mathbf{q} = \mathbf{0},
 \end{aligned} \tag{14}$$

where:

$$\begin{aligned}
 \mathbf{B}_1 &= \int_0^1 \Phi \Phi_x^T dx / \int_0^1 \Phi \Phi^T dx, \quad \mathbf{B}_2 = \int_0^1 \Phi \Phi_{xx}^T dx / \int_0^1 \Phi \Phi^T dx, \\
 \mathbf{B}_3 &= \int_0^1 \Phi \Phi_{xxx}^T dx / \int_0^1 \Phi \Phi^T dx, \quad \mathbf{B}_4 = \int_0^1 \Phi \Phi_{xxxx}^T dx / \int_0^1 \Phi \Phi^T dx,
 \end{aligned}$$

are all square matrix of n_G th order, n_G is the number of items of $\phi(x)$.

Equation (14) can be expressed in matrix form as:

$$\mathbf{u}_t = \mathbf{A} \mathbf{u}, \tag{15a}$$

where $\mathbf{u} = [\mathbf{q}^T \ \mathbf{q}_t^T \ \mathbf{q}_{tt}^T \ \mathbf{q}_{ttt}^T]^T$, \mathbf{A} is the coefficient matrix:

$$\mathbf{A} = \begin{bmatrix} \mathbf{0} & \mathbf{I} & \mathbf{0} & \mathbf{0} \\ \mathbf{0} & \mathbf{0} & \mathbf{I} & \mathbf{0} \\ \mathbf{0} & \mathbf{0} & \mathbf{0} & \mathbf{I} \\ \mathbf{A}_0 & \mathbf{A}_1 & \mathbf{A}_2 & \mathbf{A}_3 \end{bmatrix}, \tag{15b}$$

$$\begin{aligned} \mathbf{A}_0 &= -[v_t \mathbf{B}_1 + (2k_2 v_t^2 + v^2 + 1) \mathbf{B}_2 + k_1 v_t \mathbf{B}_3 + (k_4 - k_1 - k_1 v^2) \mathbf{B}_4] / k_2, \\ \mathbf{A}_1 &= -[2v \mathbf{B}_1 - 4 k_2 v v_t \mathbf{B}_2 + 2 k_1 \mathbf{B}_3] / k_2, \\ \mathbf{A}_2 &= [-\mathbf{I} - 5 k_2 v_t \mathbf{B}_1 + (k_1 + k_3 - k_2 - k_2 v^2) \mathbf{B}_2] / k_2, \\ \mathbf{A}_3 &= -2v \mathbf{B}_1, \end{aligned}$$

where $\mathbf{0}$ and \mathbf{I} are the n order zero matrix and unit matrix, respectively, and as a typical gyroscopic matrix, the translation exists for \mathbf{A} as follow:

$$\mathbf{T}^{-1} \mathbf{A} \mathbf{T} = \begin{bmatrix} \omega_1 \mathbf{J} & & \\ & \dots & \\ & & \omega_{2n} \mathbf{J} \end{bmatrix}, \quad \mathbf{J} = \begin{bmatrix} 0 & -1 \\ 1 & 0 \end{bmatrix}. \quad (16)$$

The eigenvalue ω_i of matrix \mathbf{A} are pairs of complex numbers. According to Eqs. (14) and (15), the number of items for MGM truncation must be larger than the number of natural frequency order to be calculated, therefore, n_G is determined to be 4 for the accurate calculation for the 1st and 2nd natural frequencies.

3.3. Differential quadrature method

Substituting Eq. (9) into Eq. (8) yields:

$$\begin{aligned} k_2 \phi \lambda^4 - 2 k_2 v \phi_x i \lambda^3 - [\phi + 5 k_2 v_t \phi_x - (k_1 + k_3 - k_2 - k_2 v^2) \phi_{xx}] \lambda^2 \\ + [2v \phi_x + 4 k_2 v v_t \phi_{xx} - 2 k_1 v \phi_{xxx}] i \lambda \\ + [v_t \phi + (2 k_2 v_t^2 + v^2 + 1) \phi_x + k_1 v_t \phi_{xxx} + (k_4 - k_1 - k_1 v^2) \phi_{xxxx}] = 0. \end{aligned} \quad (17)$$

According to the differential quadrature method, r th order derivative value of function $f(x)$ at point x_i can be expressed as a weighted sum of the value of all the nodes:

$$f_x^{(r)}(x_i) = \sum_{j=1}^n A_{ij}^{(r)} f_j, \quad i = 1, 2, \dots, n, \quad (18)$$

where $A_{ij}^{(r)}$ presents the r th order weight coefficients of derivative. f_j is the function value at x_j , n is the number of nodes. Therefore the first four order derivatives of mode functions of the nodes can be obtained by interpolation of the function values of the nodes as:

$$\begin{aligned} \phi_x(x_i) &= \sum_{j=1}^n A_{ij}^{(1)} \phi_j, & \phi_{xx}(x_i) &= \sum_{j=1}^n A_{ij}^{(2)} \phi_j, \\ \phi_{xxx}(x_i) &= \sum_{j=1}^n A_{ij}^{(3)} \phi_j, & \phi_{xxxx}(x_i) &= \sum_{j=1}^n A_{ij}^{(4)} \phi_j. \end{aligned} \quad (19)$$

Non-uniform node distribution is used to discretize the beam and the δ method is introduced to deal with the boundary conditions, which is to add two nodes of δ distance from the two ends and renumber the nodes:

$$\begin{aligned} x_1 = 0, \quad x_2 = \delta, \quad x_i = \frac{1}{2} \left[1 - \cos \frac{(i-1)\pi}{n-1} \right], \\ i = 3, 4, \dots, n-2, \quad x_{n-1} = 1 - \delta, \quad x_n = 1, \end{aligned} \quad (20)$$

where $\delta = 10^{-4} - 10^{-6}$ [19]. The weight coefficients of DQM can be obtained through the Lagrange

interpolation polynomial according to the interpolation principle. Expressions of the coefficients are:

$$A_{ij}^{(1)} = \begin{cases} \prod_{k=1, k \neq i}^n (x_i - x_k) / (x_i - x_j) \prod_{k=1, k \neq j}^n (x_j - x_k), & i, j = 1, 2, \dots, n, \quad i \neq j, \\ \sum_{k=1, k \neq i}^n \frac{1}{(x_i - x_k)}, & i, j = 1, 2, \dots, n, \quad i = j, \end{cases} \quad (21)$$

$$A_{ij}^{(r)} = \begin{cases} r \left(A_{ii}^{(r-1)} A_{ij}^{(1)} - \frac{A_{ij}^{(r-1)}}{(x_i - x_j)} \right), & i, j = 1, 2, \dots, n, \quad i \neq j, \\ - \sum_{k=1, k \neq i}^n A_{ik}^{(r)}, & i, j = 1, 2, \dots, n, \quad i = j. \end{cases} \quad (22)$$

Substituting Eqs. (21) and (22) into Eq. (17) yields:

$$\begin{aligned} & k_2 \phi_i \lambda^4 - 2 k_2 v i \lambda^3 \sum_{j=1}^n A_{ij}^{(1)} \phi_j \\ & + \left[2v \sum_{j=1}^n A_{ij}^{(1)} \phi_j + 4 k_2 v v_t \sum_{j=1}^n A_{ij}^{(2)} \phi_j - 2 k_1 v \sum_{j=1}^n A_{ij}^{(3)} \phi_j \right] i \lambda \\ & - \left[\phi_i + 5 k_2 v_t \sum_{j=1}^n A_{ij}^{(1)} \phi_j - (k_1 + k_3 - k_2 - k_2 v^2) \sum_{j=1}^n A_{ij}^{(2)} \phi_j \right] \lambda^2 \\ & + k_1 v_t \sum_{j=1}^n A_{ij}^{(3)} \phi_j \\ & + \left[v_t \phi_i + (2 k_2 v_t^2 + v^2 + 1) \sum_{j=1}^n A_{ij}^{(1)} \phi_j + (k_4 - k_1 - k_1 v^2) \sum_{j=1}^n A_{ij}^{(4)} \phi_j \right] = 0. \end{aligned} \quad (23)$$

Also, Eq. (23) can be translated to the matrix form as:

$$(\mathbf{B}^{(4)} \lambda^4 + \mathbf{B}^{(3)} \lambda^3 + \mathbf{B}^{(2)} \lambda^2 + \mathbf{B}^{(1)} \lambda + \mathbf{B}^{(0)}) \Phi = \mathbf{0}, \quad (24)$$

where, $\mathbf{B}^{(i)}$, $i = 0-4$ are all n th order square matrix, $\Phi = [\phi_1 \ \phi_2 \ \dots \ \phi_n]^T$ is the model function vector. Expression of elements in $\mathbf{B}^{(i)}$ are:

$$\begin{aligned} B_{ij}^{(0)} &= v_t \delta_{ij} + k_1 v_t A_{ij}^{(3)} + (2 k_2 v_t^2 + v^2 + 1) A_{ij}^{(2)} + (k_4 - k_1 - k_1 v^2) A_{ij}^{(4)}, \\ B_{ij}^{(1)} &= (2v A_{ij}^{(1)} + 4 k_2 v v_t A_{ij}^{(2)} - 2 k_1 v A_{ij}^{(3)}) i, \\ B_{ij}^{(2)} &= -\delta_{ij} - 5 k_2 v_t A_{ij}^{(1)} + (k_1 + k_3 - k_2 - k_2 v^2) A_{ij}^{(2)}, \\ B_{ij}^{(3)} &= -k_2 v i A_{ij}^{(1)}, \\ B_{ij}^{(4)} &= k_2 \delta_{ij}, \end{aligned} \quad (25)$$

where $\delta_{ij} = \begin{cases} 0, & i \neq j, \\ 1, & i = j. \end{cases}$

For nodes at the two ends, i.e. $i, j = 1, 2, n-1, n$, since that the introduction of the

time-varying items, not only elements in $\mathbf{B}^{(0)}$ are affected, but also in $\mathbf{B}^{(1)}$ and $\mathbf{B}^{(2)}$. According to Table A-1, elements in $\mathbf{B}^{(0)}$, $\mathbf{B}^{(1)}$ and $\mathbf{B}^{(2)}$ are improved after introducing the boundary conditions with the axial compressive load. The DQM form of boundary conditions are shown in Table A-4 in Appendix.

Then solving of Eq. (24) is transformed to the generalized eigenvalue problem. The eigenvalues of Eq. (24) represents the dimensionless natural frequency of beam system.

4. Verification and discussion

An uniform thin-walled beam with rectangular cross-section is used for solutions and discussions. The beam's configuration and material parameters are: $L = 1\text{ m}$, $A = 0.02\text{ m}^2$, $I = 1.67 \times 10^{-5}\text{ m}^4$, $A = 0.02\text{ m}^2$, $E = 100\text{ GPa}$, $\mu = 0.3$, $\rho = 1.89 \times 10^3\text{ kg/m}^3$.

4.1. Verification of numerical methods

The beam in engineering can be classified into two types according to the positive definiteness property of boundary condition as mentioned before. Take one from each type of boundary conditions for example, i.e. the pinned-pinned beam for PDS, and the free-free beam for PSDS. The numerical results of MGM and DQM are compared with the analytical ones under $v = v_t = 0$, as shown in Fig. 5.

Fig. 5 shows that numerical results of MGM and DQM both agree with analytical results in the range of $k_n = 0.01-1$, thus these two numerical methods are both applicable for different boundary conditions. Fig. 6 shows the results of DQM and MGM for the pinned-pinned beam in the range of $v = 0-4$, the results for 1st and 2nd natural frequencies of these two methods are shown to agree with each other well. The DQM is used in the following numerical analysis while the number of nodes n is set to 13 for convergence.

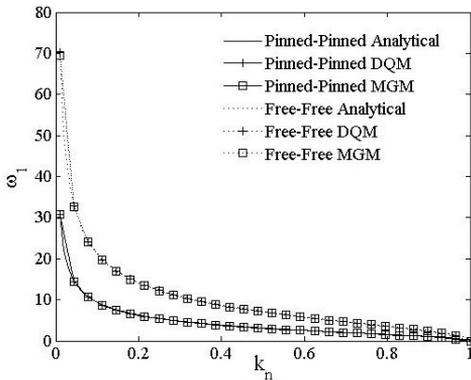


Fig. 5. Comparison of DQM, MGM and analytical solutions ω_1 vs. k_n

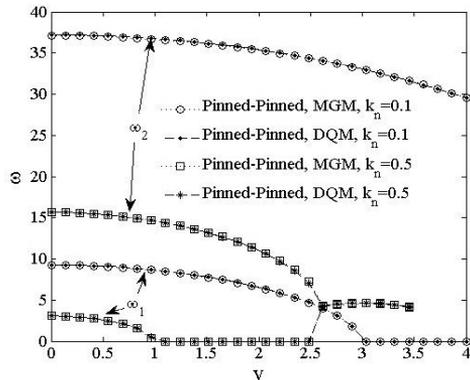


Fig. 6. Comparison of DQM, MGM $\omega_{1,2}$ vs. v

4.2. Effect of axial compressive load

Fig. 7 indicates the variation of the 1st dimensionless complex frequency of PDS with axial compressive load. The complex frequencies of PDS are all real numbers with imaginary parts equal to 0 when axially moving speed is 0. The real part decreases rapidly in the range of $k_n = 0.01-0.2$ and slowly when $k_n = 0.2-1$. When the load reaches to the critical load as $k_n = 1$, $Re(\omega_1)$ reduce to 0, which indicates the critical load factor $k_{ncr} = 1$ when $v = 0$. But for $v = 1$, the imaginary part of clamped-free beam is negative. With increasing moving speed for all boundary conditions the real part of the 1st frequency reduces to zero and it becomes unstable by

the divergence instability of their imaginary parts when the load reaches the critical loads, which decreases to around half of the value of static beam.

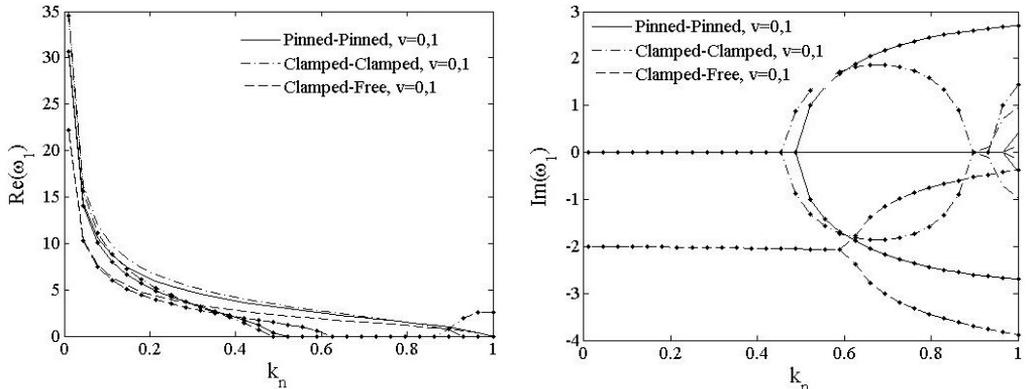


Fig. 7. The 1st dimensionless complex frequency of PDS vs. load $v_t = 0, v = 0, 1$.
 $v = 0$ --- ; $v = 1$ - · -

Fig. 8 indicates the variation of the 1st dimensionless complex frequency of PSDS with axial compressive load. As the cases in PDS, the frequency reduces with the increasing of moving speed. When $v = 1$, the critical loads of free-free and pinned-free beams reduce to $0.8N_{Cr}$ and $0.6N_{Cr}$, respectively. The imaginary parts $Im(\omega_1)$ of those two conditions keep 0 before critical load when $v = 0$. When $v = 1$, it branches for free-free beam and the bifurcation point is the critical load point, but $Im(\omega_1)$ of pinned-free beam keeps rising with increasing load and it intersects the horizontal axis at the critical load point.

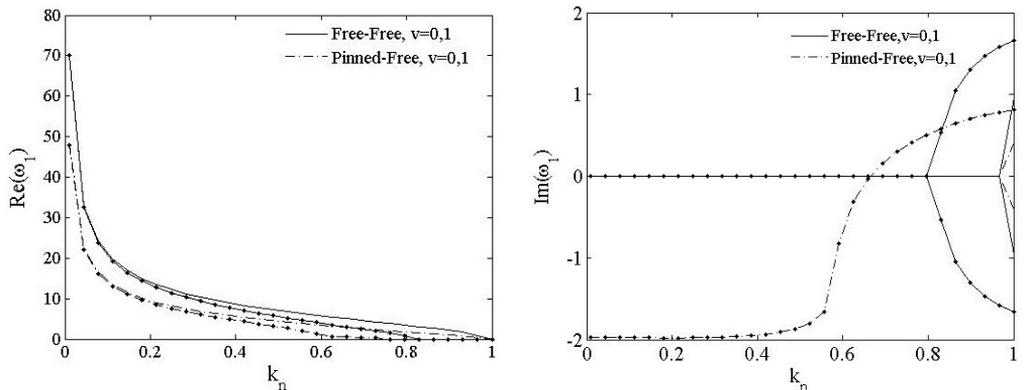


Fig. 8. The 1st dimensionless complex frequency of PSDB vs. load $v_t = 0, v = 0, 1$.
 $v = 0$ --- ; $v = 1$ - · -

Fig. 9 indicates the variation of the imaginary parts of the fundamental modes of PSDS with axial compressive load. There exists two fundamental modes for free-free beam, one for in plane translational motion and the other for in plane rotation, while the pinned-free beam only has one for rotation. The real part of fundamental mode keep 0 for the two boundary conditions, and only the imaginary parts need to be discussed. For free-free beam, the imaginary frequency of the first mode of translational motion keeps 0, while that of the second mode of in plane rotation has slight increasing with k_n for $v = 0$ and has an obvious increasing when $k_n > 0.5$ for $v = 1$. For pinned-free beam, the imaginary frequency of in plane rotation keeps increasing in a small range with respect to k_n for $v = 0$, but it has a complex variation of

bifurcation-combination-rebifurcation phenomenon with the increasing of k_n for $\nu = 1$.

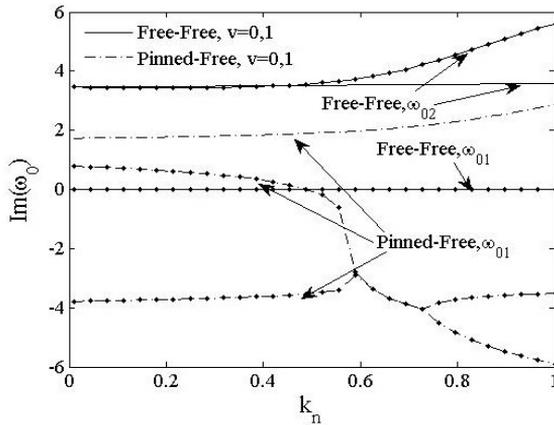


Fig. 9. Imaginary part of fundamental frequencies of PSDB vs. load $\nu_t = 0$, $\nu = 0, 1$.
 $\nu = 0$ — — ; $\nu = 1$ - - -

4.3. Effect of axially moving

Figs. 10-12 show the variations of the first three dimensionless complex frequencies of PDSs boundary cases with axially moving speed when $\nu_t = 0$, $k_n = 0.1, 0.5$. For all the three boundary cases, the real parts of ω , i.e. $Re(\omega_1)$ decrease with the increasing of speed. The $Re(\omega_1)$ reduces to 0 when the moving speed rises to the critical value ν_{cr} and the supercritical value of speed ν_{scr} . The ν_{cr} of three cases are 6.5, 2.8 and 3.9 when $k_n = 0.1$, the values decrease to 1.0, 1.1 and 1.3 when $k_n = 0.5$, which indicates that axial the compressive axial load affects the critical moving speed a lot. For $\nu_{cr} < \nu < \nu_{scr}$, the 2nd and 3rd mode remain stable, but 1st mode is unstable with divergence instability. Special attention should be paid to the left parts of Fig. 10 for $k_n = 0.5$ and Fig. 11 for $k_n = 0.1, 0.5$. The 1st order regains stability when ν reaches ν_{scr} , after that, the real parts of 1st and 2nd complex frequencies merge to each other while the imaginary parts bifurcate with positive and negative values, which indicates that the two order modes couple with each other and the beam undergoes coupled-mode flutter. For the clamped-free beam shown in Fig. 12, the real parts of 1st and 2nd complex frequencies get close but not merge together when ν reaches ν_{scr} , and The $Im(\omega)$ of clamped-free beam keep negative as moving speed increasing and is different from the other two PDSs as a result of its asymmetric boundary condition.

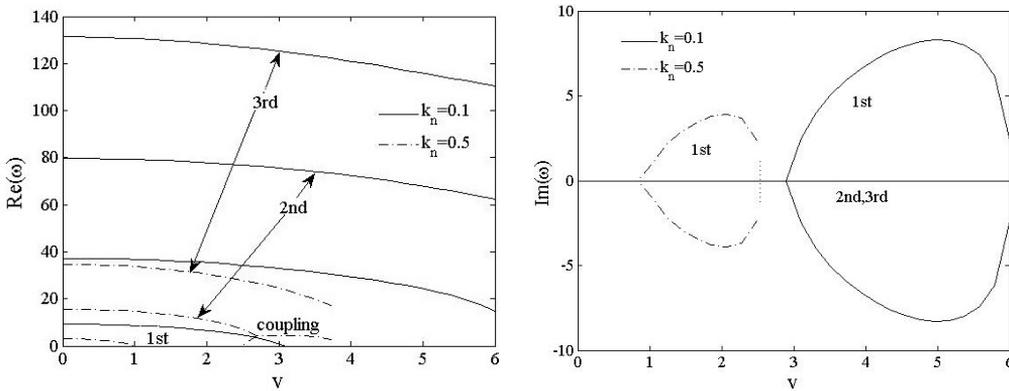


Fig. 10. The 1st, 2nd and 3rd dimensionless complex frequencies of pinned-pinned vs. speed $\nu_t = 0$, $k_n = 0.1, 0.5$

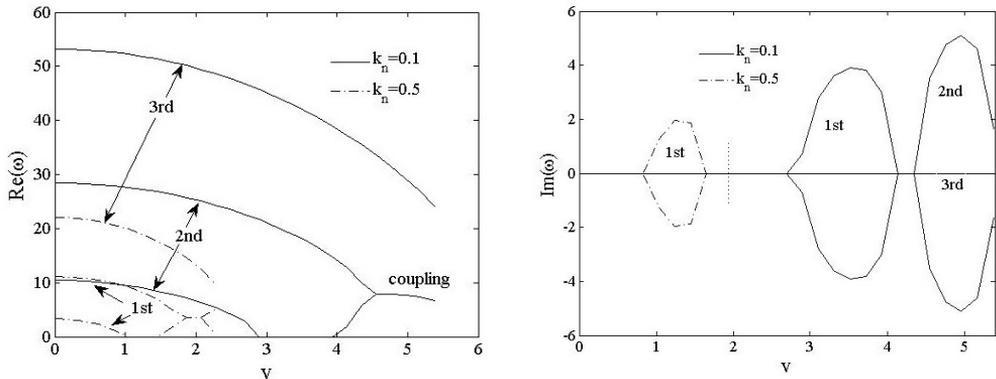


Fig. 11. The 1st, 2nd and 3rd dimensionless complex frequencies of clamped-clamped vs. speed $v_t = 0$, $k_n = 0.1, 0.5$

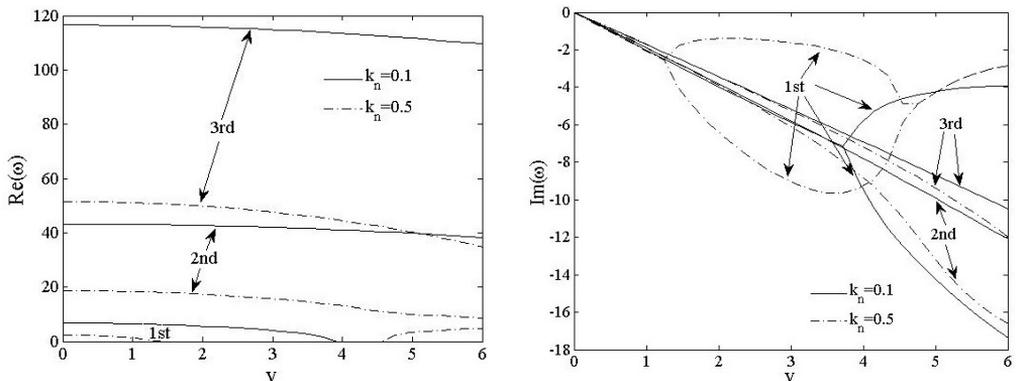


Fig. 12. The 1st, 2nd and 3rd dimensionless complex frequencies of clamped-free vs. speed $v_t = 0$, $k_n = 0.1, 0.5$

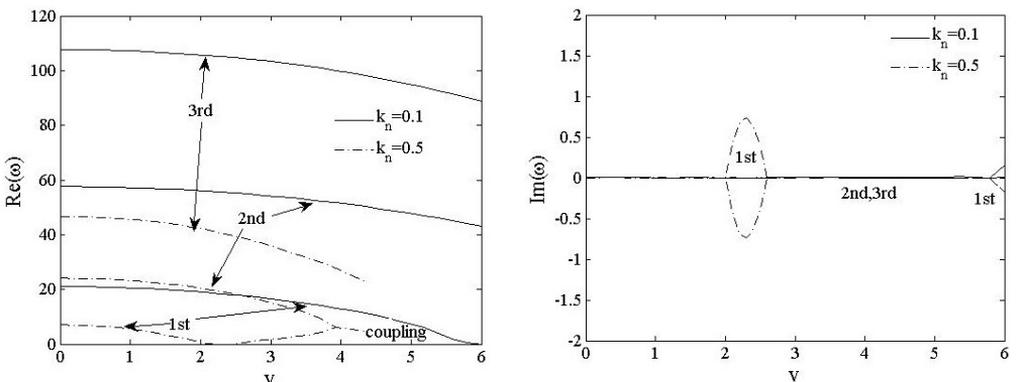


Fig. 13. The 1st, 2nd and 3rd dimensionless complex frequencies of free-free vs. speed $v_t = 0$, $k_n = 0.1, 0.5$

Figs. 13 and 14 show the variations of the first three order dimensionless complex frequencies of PSDSs boundary cases with axially moving speed when $v_t = 0$, $k_n = 0.1, 0.5$. For free-free beam shown in Fig. 13, v_{cr} reduces from 5.9 to 2.2 when the compressive axial load rises from $0.1N_{cr}$ to $0.5N_{cr}$. For $k_n = 0.5$, when $2.2 < v < 2.6$, the positive parts of 1st complex frequency keeps stable with value 0 while imaginary parts branches. When $v > 2.6$, $Re(\omega_1)$ reappears and rises up to couple with $Re(\omega_2)$ at $v = 3.9$. $Re(\omega_3)$ decreases with load increasing while $Im(\omega_2)$

and $Im(\omega_3)$ both remains 0.

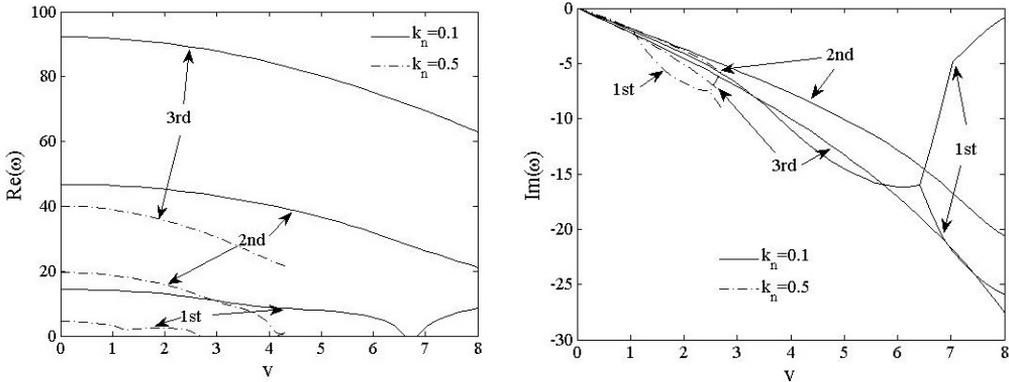


Fig. 14. The 1st, 2nd and 3rd dimensionless complex frequencies of pinned-free vs. speed $v_t = 0, k_n = 0.1, 0.5$

For the pinned-free beam shown in Fig. 14, v_{cr} is 6.6 under the load of $0.1N_{cr}$, and $Im(\omega_1)$ branches and is asymmetric. The beam undergoes supercritical state when v reaches to 6.9, 1st and 2nd frequencies are close to each other but do not couple. v_{cr} reduces to 2.6 when the compressive load increases to $0.5N_{cr}$. For all cases, $Re(\omega_2)$ and $Re(\omega_3)$ are positive while $Im(\omega_2)$ and $Im(\omega_3)$ are both negative and decrease with increasing speed.

Fig. 15 show the variations of the fundamental modes of PSDSs boundary cases with axially moving speed when $v_t = 0, k_n = 0.1, 0.5$. The 1st fundamental frequency of free-free beam remains 0, while the 2nd order frequency decreases slowly with increasing moving speed but rises sharply when v near to v_{cr} for $k_n = 0.1$. The $Im(\omega_0)$ branches when $v > 0$ and merges at $v = 4.6$. The law of $Im(\omega_0)$ for those two boundary cases are similar, when $k_n = 0.5$ the turning point moves forward.

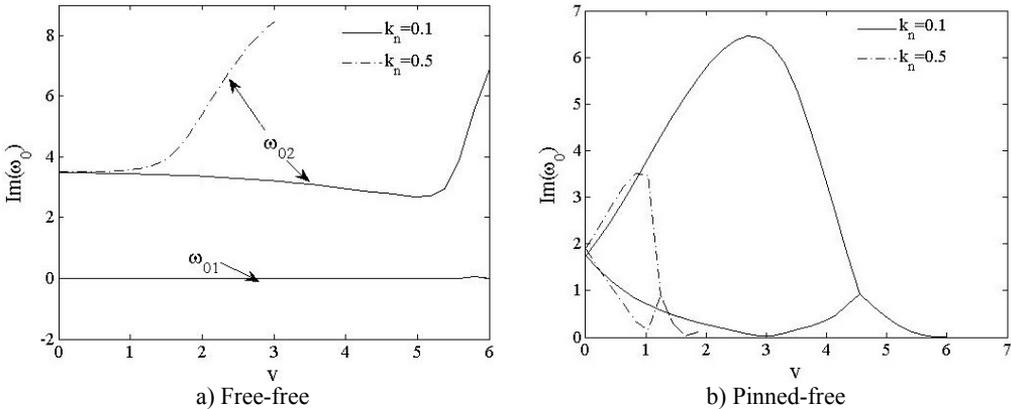


Fig. 15. Fundamental frequencies of PSDB vs. speed $v_t = 0, k_n = 0.1, 0.5$

Figs. 16 and 17 show the variations of the first three dimensionless frequencies of pinned-pinned and pinned-free beams with axially moving acceleration when $k_n = 0.1, 0.5, v = 0, 3$. Fig. 16 indicates that the real parts of each order frequencies rise with the increasing moving acceleration when axial load is $0.1N_{cr}$. When the load increase to $0.5N_{cr}$, $Re(\omega_1)$ turns to decrease for $v_t > 400$ and reaches 0 when $v_t = 670$, while the 2nd and 3rd modes couple each other which indicates that the beam undergoes coupled-mode flutter. For pinned-free beam, fundamental mode merges with the 1st elastic mode when $v_t = 530$, and the 2nd and 3rd modes

couple at $v_t = 670$. Thus, the relationship between elastic modes and axially moving acceleration is complex under a nonlinear coupling with the axial compressive load.

From Fig. 17, one can see that the real parts of each order rise with increasing moving acceleration. For $v_t > 500$ $Re(\omega_1)$ of pinned-free beam is larger at $v = 3$ than $v = 1$, which indicates that the effect of coupling speed and acceleration is not always linear.

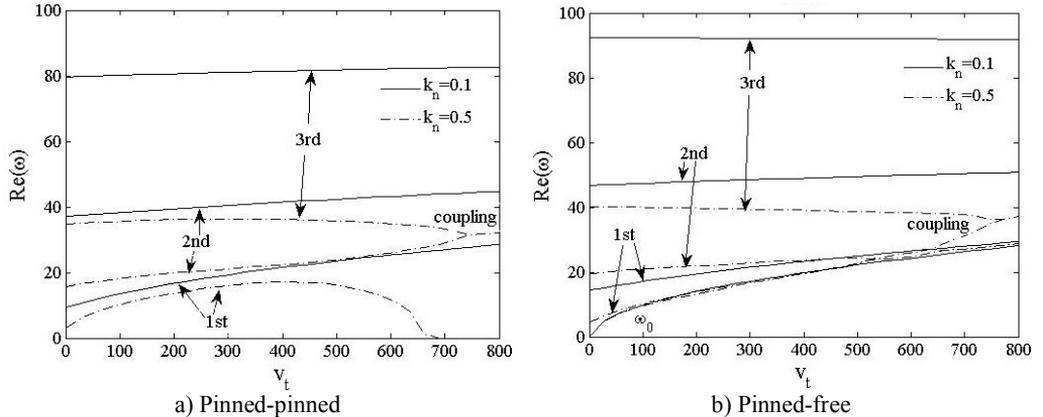


Fig. 16. Real part of the first three frequencies vs. acceleration $k_n = 0.1, 0.5, v = 0$

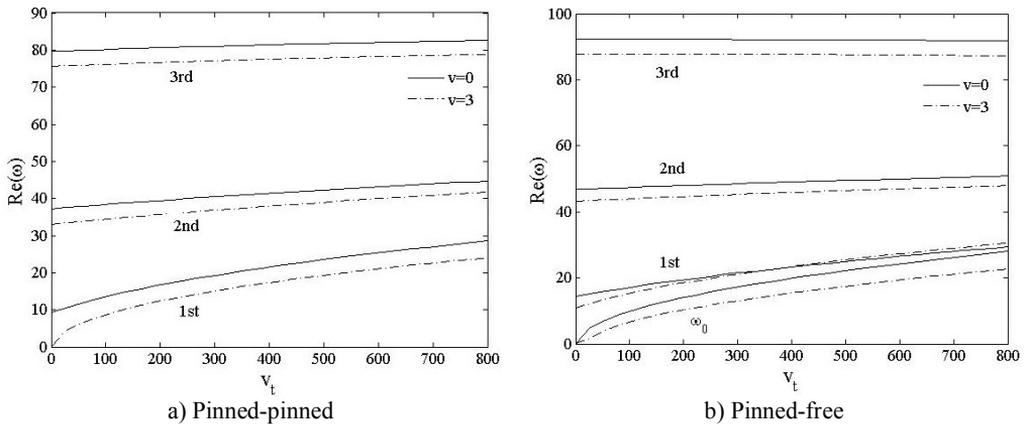


Fig. 17. Real part of the first three frequencies vs. acceleration $k_n = 0.1, v = 0, 1$

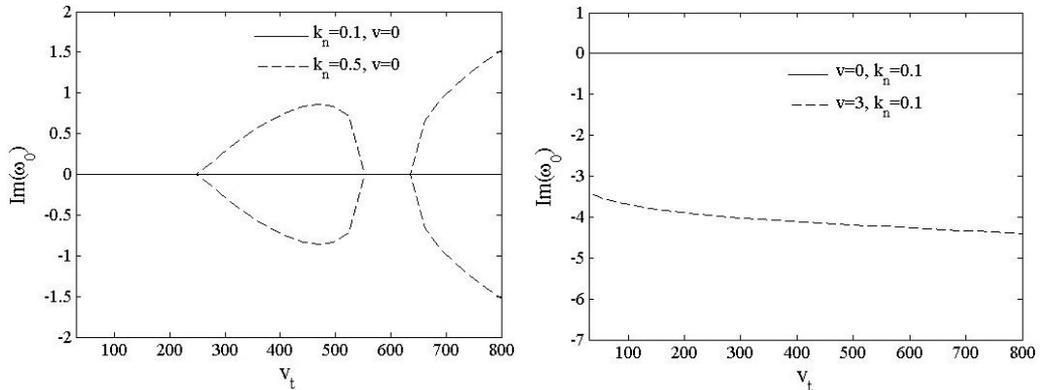


Fig. 18. Fundamental frequencies of pinned-free beam vs. acceleration $k_n = 0.1, 0.5, v = 0, 3$

Fig. 18 shows the variations of the fundamental frequencies of pinned-free beam with axially moving acceleration when $k_n = 0.1, 0.5, v = 0$ and $v = 0, 3, k_n = 0.1$. The $Im(\omega_0)$ branches when $v_t = 240$ and this point agrees with $Re(\omega_1)$. The $Im(\omega_0)$ remains 0 when $v = 0$ and decreases with the increasing of acceleration when $v > 0$.

5. The critical loads and speeds

When the first elastic frequency w_1 reduces to zero, the beam reaches a critical state, and Eq. (8) becomes a time-independent equilibrium linear equation:

$$v_t w_x + (2k_2 v_t^2 + v^2 + 1)w_{xx} + k_1 v_t w_{xxx} + (k_4 - k_1 - k_1 v^2)w_{xxxx} = 0. \tag{26}$$

It can be simplified without taking into account of axially acceleration i.e. $v_t = 0$:

$$w_{xxxx} + \frac{(v^2 + 1)}{(k_4 - k_1 - k_1 v^2)} w_{xx} = 0. \tag{27}$$

The roots of the characteristic equation of Eq. (27) is:

$$r_1 = r_2 = 0, \quad r_{3,4} = \pm \sqrt{\frac{(v^2 + 1)}{(k_4 - k_1 - k_1 v^2)}}. \tag{28}$$

And the general solution of Eq. (27) is:

$$w(x) = C_1 + C_2 x + C_3 \sin \beta x + C_4 \cos \beta x. \tag{29}$$

Equation (29) contains two parts, the first part $C_1 + C_2 x$ denotes two types of foundational motion, i.e. lateral rigid motion, and the in-plane rotation. For PDS, the two foundational motions are limited. For PSDS, free-free beam have both foundational motions and pinned-free beam only has in-plane rotation. The second part denotes elastic modes.

Equation (27) can be expressed to a similar form of Eq. (10), where $\alpha_1^2 = \frac{v^2+1}{k_4-k_1-k_1v^2}$, $\alpha_2 = 0$. Then n th eigenvalue can be obtained α_n are obtained: $\alpha_n = n\pi$, for pinned-pinned, pinned-free and free-free beams, $\alpha_n = 2n\pi$ for clamped-clamped beam, $\alpha_n = (2n - 1)\pi/2$ for clamped-free beams. By defining a new parameters $k_6 = \frac{EI^2}{\kappa GA^2 L^4}$, then $k_1 = k_6 \mu k_n$, $k_4 = 1/\mu k_n$. Taking these items into the expression of α_1 , the relationship between critical moving speed and axial compressive load can be obtained as:

$$v_{cr} = \sqrt{\frac{\alpha_1^2(1/\mu k_n - k_6)}{1 + \alpha_1^2 k_6}}. \tag{30}$$

And the variation of critical load to moving speed can also be obtained at the same time.

Fig. 19 shows the relationship between critical load and speed of clamped-free beam. It indicates that with the effect of axially moving and axial compressive load, the beam reaches critical state more easily. The critical value of speed or load can be approximately predicted from these results.

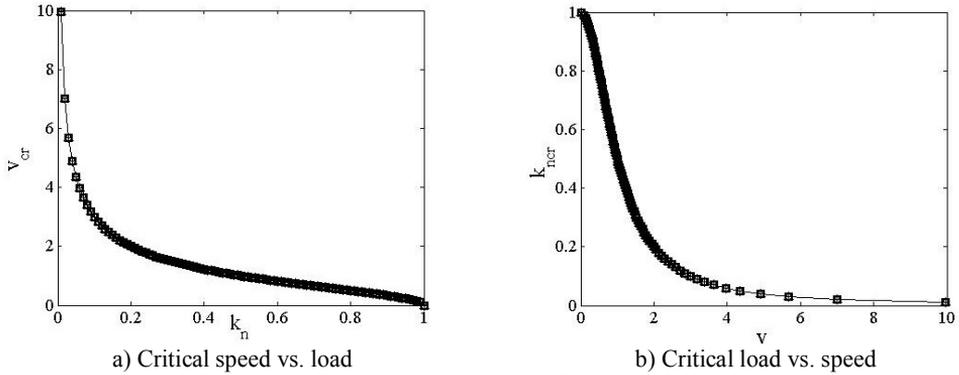


Fig. 19. Critical load and speed of clamped-free beam

6. Conclusions

In this paper, the effect of compressive load on complex frequencies and stabilities of axially moving Timoshenko beam with different supports are studied, expressions of the critical load and speed for various boundary conditions are determined. Results of the study can be summarized as follows:

The real part of first three complex frequencies of five different boundary conditions decrease with the increasing compressive load and axially moving speed. The first two order modes of PDS conditions undergo coupled-mode flutter with increasing critical moving speed by divergent instability, while those of PSDS get closer but not merge together also. Rigid mode of PSDS undergoes instability with the increasing of compressive load and speed. The moving acceleration causes increasing of elastic and foundational frequencies.

Critical moving speed decreases with the increasing of the axial load, the critical load decreases with the increasing of axially moving speed, the coupling of these two effects causes the beam to reach instability state more easily.

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Appendix

Table A-1. Dimensionless form of different boundary conditions and critical loads

Boundary conditions	Original form	Dimensionless form	N_{cr}
Pinned-Pinned 	$X = 0, L$ $\{W(X) = 0$ $\{M(X) = 0$	$x = 0, 1$ $w = 0$ $(1 - k_5 v^2 - k_5)w_{xx}$ $-(k_5 v_t + 2 k_5 v i \lambda)w_x = 0$	$\frac{\pi^2 EI}{L^2}$
Clamped-Clamped 	$X = 0, L$ $\{W(X) = 0$ $\{\theta(X) = 0$	$x = 0, 1$ $w = 0$ $w_x = 0$	$\frac{4\pi^2 EI}{L^2}$
Free-Free 	$X = 0, L$ $\{M(X) = 0$ $\{Q(X) + NW_X = 0$	$x = 0, 1$ $(1 - k_5 v^2 - k_5)w_{xx} + k_5 \lambda^2 w$ $-(k_5 v_t + 2 k_5 v i \lambda)w_x = 0$ $(k_1 v^2 + k_1 - k_4)w_{xxx} - k_1 \lambda^2 w_x$ $+(k_1 v_t + 2 k_1 v i \lambda)w_{xx} = 0$	$* \frac{\pi^2 EI}{L^2}$
Pinned-Free 	$X = 0$ $\{W(X) = 0$ $\{M(X) = 0$ $X = L$ $\{M(X) = 0$ $\{Q(X) + NW_X = 0$	$x = 0$ $w = 0$ $(1 - k_5 v^2 - k_5)w_{xx}$ $-(k_5 v_t + 2 k_5 v i \lambda)w_x = 0$ $x = 1$ $(1 - k_5 v^2 - k_5)w_{xx} + k_5 \lambda^2 w$ $-(k_5 v_t + 2 k_5 v i \lambda)w_x = 0$ $(k_1 v^2 + k_1 - k_4)w_{xxx} - k_1 \lambda^2 w_x$ $+(k_1 v_t + 2 k_1 v i \lambda)w_{xx} = 0$	$* \frac{\pi^2 EI}{L^2}$
Clamped-Free 	$X = 0$ $\{W(X) = 0$ $\{\theta(X) = 0$ $X = L$ $\{M(X) = 0$ $\{Q(X) + NW_X = 0$	$x = 0, 1$ $w = 0$ $w_x = 0$ $x = 1$ $(1 - k_5 v^2 - k_5)w_{xx} + k_5 \lambda^2 w$ $-(k_5 v_t + 2 k_5 v i \lambda)w_x = 0$ $(k_1 v^2 + k_1 - k_4)w_{xxx} - k_1 \lambda^2 w_x$ $+(k_1 v_t + 2 k_1 v i \lambda)w_{xx} = 0$	$\frac{\pi^2 EI}{4L^2}$

*critical load of PSDS for reference

Table A-2. Characteristics equations of different boundary conditions [14]

Boundary cases	Characteristic equation
Pinned-Pinned	$\sin \alpha_1 = 0$
Clamped-Clamped	$2\lambda_i^4 - k_s^2 \sinh \alpha_2 \sin \alpha_1 - 2\lambda_i^2 \cosh \alpha_2 \cos \alpha_1 = 0$
Free-Free	$2\lambda_i^6 [1 - \cosh \alpha_2 \cos \alpha_1] + k_s^2 (k_s^4 + 3\lambda_i^4) \sinh \alpha_2 \sin \alpha_1 = 0$
Pinned-Free	$\alpha_1^3 \cosh \alpha_2 \cos \alpha_1 - \alpha_2^3 \sinh \alpha_2 \sin \alpha_1 = 0$
Clamped-Free	$2\lambda_i^4 - \lambda_i^2 k_s^2 \sinh \alpha_2 \sin \alpha_1 + (k_s^4 + 2\lambda_i^4) \cosh \alpha_2 \cos \alpha_1 = 0$

Table A-3. Mode shape functions of different boundary conditions [14]

Boundary cases	Mode shape function $\phi(x)$
Pinned-Pinned	$\sin \alpha_1 x$
Clamped-Clamped	$\sinh \alpha_2 x + \frac{\alpha_2 \sin \alpha_1 - \alpha_1 \sinh \alpha_2}{\alpha_1 (\cosh \alpha_2 - \cos \alpha_1)} \cosh \alpha_2 x - \frac{\alpha_1}{\alpha_2} \sin \alpha_1 x - \frac{\alpha_2 \sin \alpha_1 - \alpha_1 \sinh \alpha_2}{\alpha_1 (\cosh \alpha_2 - \cos \alpha_1)} \cos \alpha_1 x$
Free-Free	$\sinh \alpha_2 x + \frac{\alpha_2^3 (\cosh \alpha_2 - \cos \alpha_1)}{\alpha_1^3 \sinh \alpha_2 + \alpha_2^3 \sin \alpha_1} \cosh \alpha_2 x - \frac{\alpha_2}{\alpha_1} \sin \alpha_1 x - \frac{\alpha_1^2 \alpha_2 (\cosh \alpha_2 - \cos \alpha_1)}{\alpha_2^3 \sinh \alpha_2 + \alpha_2^3 \sin \alpha_1} \cos \alpha_1 x$
Pinned-Free	$\frac{\alpha_2^3 \sinh \alpha_2}{\alpha_1^3 \sin \alpha_1} \sin \alpha_1 x$
Clamped-Free	$\sinh \alpha_2 x + \frac{\alpha_2^2 \sinh \alpha_1 + \alpha_1 \alpha_2 \sin \alpha_1}{\alpha_1^2 \cosh \alpha_2 + \alpha_2^2 \cos \alpha_1} \cosh \alpha_2 x - \frac{\alpha_2}{\alpha_1} \sin \alpha_1 x + \frac{\alpha_2^2 \sinh \alpha_1 + \alpha_1 \alpha_2 \sin \alpha_1}{\alpha_1^2 \cosh \alpha_2 + \alpha_2^2 \cos \alpha_1} \cos \alpha_2 x$

Table A-4. DQM form of different boundary conditions

Pinned-Pinned	$(k = 1, n; j = 2, \dots, n - 1)$ $B_{kk}^{(0)} = \delta_{kj}$	$(k = 2, n - 1; j = 1, 2, \dots, n - 1, n)$ $B_{kj}^{(0)} = (1 - k_5 v^2 - k_5) A_{kj}^{(2)} - k_5 v_t \delta_{kj}$
Clamped-Clamped	$(k = 1, n; j = 2, \dots, n - 1)$ $B_{kk}^{(0)} = \delta_{kj}$	$(k = 2, n - 1; j = 1, 2, \dots, n - 1, n)$ $B_{kj}^{(0)} = A_{kj}^{(1)}$
Free-Free	$(k = 1, n; j = 2, \dots, n - 1)$ $B_{kj}^{(0)} = (k_1 v^2 + k_1 - k_4) A_{kj}^{(3)} + k_1 v_t A_{kj}^{(2)}$ $B_{kj}^{(1)} = 2k_1 v_i A_{kj}^{(2)}$ $B_{kj}^{(2)} = -k_1 v_i A_{kj}^{(1)}$	$(k = 2, n - 1; j = 1, 2, \dots, n - 1, n)$ $B_{kj}^{(0)} = (1 - k_5 v^2 - k_5) A_{kj}^{(2)} + k_5 v_t A_{kj}^{(1)}$ $B_{kj}^{(1)} = 2k_5 v_i A_{kj}^{(2)}$ $B_{kj}^{(2)} = k_5 A_{kk}^{(1)} \delta_{kj}$ $B_{kj}^{(m)} = 0, (k, j = 1, 2, \dots, n - 1, n), (m = 3, 4)$
Pinned-Free	$(k = 1; j = 2, \dots, n - 1)$ $B_{kk}^{(0)} = \delta_{kj}$ $(k = n; j = 2, \dots, n - 1)$ $B_{kj}^{(0)} = (k_1 v^2 + k_1 - k_4) A_{kj}^{(3)} + k_1 v_t A_{kj}^{(2)}$ $B_{kj}^{(1)} = 2k_1 v_i A_{kj}^{(2)}$ $B_{kj}^{(2)} = -k_1 v_i A_{kj}^{(1)}$	$(k = 2; j = 1, 2, \dots, n - 1, n)$ $B_{kj}^{(0)} = (1 - k_5 v^2 - k_5) A_{kj}^{(2)} - k_5 v_t \delta_{kj}$ $(k = n - 1; j = 1, 2, \dots, n - 1, n)$ $B_{kj}^{(0)} = (1 - k_5 v^2 - k_5) A_{kj}^{(2)} + k_5 v_t A_{kj}^{(1)}$ $B_{kj}^{(1)} = 2k_5 v_i A_{kj}^{(2)}$ $B_{kj}^{(2)} = k_5 A_{kk}^{(1)} \delta_{kj}$ $B_{kj}^{(m)} = 0, (k, j = 1, 2, \dots, n - 1, n), (m = 3, 4)$
Clamped-Free	$(k = 1; j = 2, \dots, n - 1)$ $B_{kk}^{(0)} = \delta_{kj}$ $(k = n; j = 2, \dots, n - 1)$ $B_{kj}^{(0)} = (k_1 v^2 + k_1 - k_4) A_{kj}^{(3)} + k_1 v_t A_{kj}^{(2)}$ $B_{kj}^{(1)} = 2k_1 v_i A_{kj}^{(2)}$ $B_{kj}^{(2)} = -k_1 v_i A_{kj}^{(1)}$	$(k = n - 1; j = 1, 2, \dots, n - 1, n)$ $B_{kj}^{(0)} = A_{kj}^{(1)}$ $(k = n - 1; j = 1, 2, \dots, n - 1, n)$ $B_{kj}^{(0)} = (1 - k_5 v^2 - k_5) A_{kj}^{(2)} + k_5 v_t A_{kj}^{(1)}$ $B_{kj}^{(1)} = 2k_5 v_i A_{kj}^{(2)}$ $B_{kj}^{(2)} = k_5 A_{kk}^{(1)} \delta_{kj}$ $B_{kj}^{(m)} = 0, (k, j = 1, 2, \dots, n - 1, n), (m = 3, 4)$